

This Page Is Inserted by IFW Operations
and is not a part of the Official Record

BEST AVAILABLE IMAGES

Defective images within this document are accurate representations of the original documents submitted by the applicant.

Defects in the images may include (but are not limited to):

- BLACK BORDERS
- TEXT CUT OFF AT TOP, BOTTOM OR SIDES
- FADED TEXT
- ILLEGIBLE TEXT
- SKEWED/SLANTED IMAGES
- COLORED PHOTOS
- BLACK OR VERY BLACK AND WHITE DARK PHOTOS
- GRAY SCALE DOCUMENTS

IMAGES ARE BEST AVAILABLE COPY.

**As rescanning documents *will not* correct images,
please do not report the images to the
Image Problem Mailbox.**

PCT

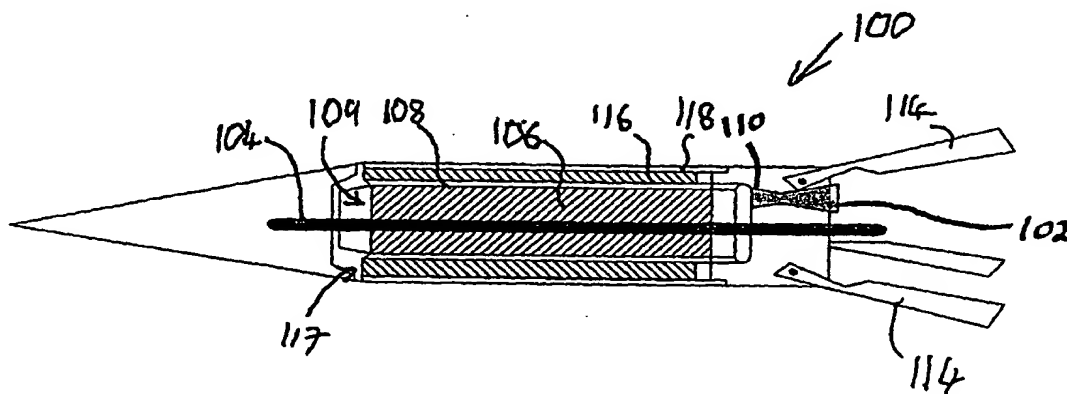
WORLD INTELLECTUAL PROPERTY ORGANIZATION
International Bureau



INTERNATIONAL APPLICATION PUBLISHED UNDER THE PATENT COOPERATION TREATY (PCT)

<p>(51) International Patent Classification 6 : F42B 15/10</p>	<p>A1</p>	<p>(11) International Publication Number: WO 00/58684 (43) International Publication Date: 5 October 2000 (05.10.00)</p>
<p>(21) International Application Number: PCT/IL99/00121 (22) International Filing Date: 25 March 1999 (25.03.99) (71) Applicant (for all designated States except US): STATE OF ISRAEL - MINISTRY OF DEFENSE RAFAEL - ARMAMENT DEVELOPMENT AUTHORITY [IL/IL]; P.O. Box 2250, 31021 Haifa (IL). (72) Inventor; and (75) Inventor/Applicant (for US only): ROTKOPF, Menachem [IL/IL]; Greenberg Street 21, 34754 Haifa (IL). (74) Agent: FRIEDMAN, Mark, M.; Beit Samueloff, Haomanim 7, 67897 Tel Aviv (IL).</p>		<p>(81) Designated States: AL, AM, AT, AU, AZ, BA, BB, BG, BR, BY, CA, CH, CN, CU, CZ, DE, DK, EE, ES, FI, GB, GD, GE, GH, GM, HR, HU, ID, IL, IN, IS, JP, KE, KG, KP, KR, KZ, LC, LK, LR, LS, LT, LU, LV, MD, MG, MK, MN, MW, MX, NO, NZ, PL, PT, RO, RU, SD, SE, SG, SI, SK, SL, TJ, TM, TR, TT, UA, UG, US, UZ, VN, YU, ZW, ARIPO patent (GH, GM, KE, LS, MW, SD, SL, SZ, UG, ZW), Eurasian patent (AM, AZ, BY, KG, KZ, MD, RU, TJ, TM), European patent (AT, BE, CH, CY, DE, DK, ES, FI, FR, GB, GR, IE, IT, LU, MC, NL, PT, SE), OAPI patent (BF, BJ, CF, CG, CI, CM, GA, GN, GW, ML, MR, NE, SN, TD, TG).</p> <p>Published With international search report.</p>

(54) Title: AN ARMOR PIERCING PROJECTILE



(57) Abstract

A projectile (100) for piercing armor including a cruise propellant (116) for maintaining a cruise velocity of the projectile. The projectile (100) also includes an acceleration rocket motor (109) activated after launch for accelerating the projectile (100) from the cruise velocity to a penetration velocity, in a final stage of flight of the projectile.

FOR THE PURPOSES OF INFORMATION ONLY

Codes used to identify States party to the PCT on the front pages of pamphlets publishing international applications under the PCT.

AL	Albania	ES	Spain	LS	Lesotho	SI	Slovenia
AM	Armenia	FI	Finland	LT	Lithuania	SK	Slovakia
AT	Austria	FR	France	LU	Luxembourg	SN	Senegal
AU	Australia	GA	Gabon	LV	Latvia	SZ	Swaziland
AZ	Azerbaijan	GB	United Kingdom	MC	Monaco	TD	Chad
BA	Bosnia and Herzegovina	GE	Georgia	MD	Republic of Moldova	TG	Togo
BB	Barbados	GH	Ghana	MG	Madagascar	TJ	Tajikistan
BE	Belgium	GN	Guinea	MK	The former Yugoslav Republic of Macedonia	TM	Turkmenistan
BF	Burkina Faso	GR	Greece			TR	Turkey
BG	Bulgaria	HU	Hungary	ML	Mali	TT	Trinidad and Tobago
BJ	Benin	IE	Ireland	MN	Mongolia	UA	Ukraine
BR	Brazil	IL	Israel	MR	Mauritania	UG	Uganda
BY	Belarus	IS	Iceland	MW	Malawi	US	United States of America
CA	Canada	IT	Italy	MX	Mexico	UZ	Uzbekistan
CF	Central African Republic	JP	Japan	NE	Niger	VN	Viet Nam
CG	Congo	KE	Kenya	NL	Netherlands	YU	Yugoslavia
CH	Switzerland	KG	Kyrgyzstan	NO	Norway	ZW	Zimbabwe
CI	Côte d'Ivoire	KP	Democratic People's Republic of Korea	NZ	New Zealand		
CM	Cameroon			PL	Poland		
CN	China	KR	Republic of Korea	PT	Portugal		
CU	Cuba	KZ	Kazakstan	RO	Romania		
CZ	Czech Republic	LC	Saint Lucia	RU	Russian Federation		
DE	Germany	LI	Liechtenstein	SD	Sudan		
DK	Denmark	LK	Sri Lanka	SE	Sweden		
EE	Estonia	LR	Liberia	SG	Singapore		

AN ARMOR PIERCING PROJECTILE

FIELD AND BACKGROUND OF THE INVENTION

5 The present invention relates to a method and apparatus for penetrating armor and, more particularly, to an armor piercing projectile.

 The use of armor to protect a combatant is wide spread on the modern battle field. An armored battle field vehicle, such as a tank, is not only heavily armed, its armor protects the vehicle's crew from exposure to enemy forces.

10 Such armored vehicles pose a high degree of threat to any attacking force. Furthermore, an active protection is often used by armored vehicles to provide further protection. Namely, shields containing water, explosives and a combination thereof are placed on an exterior surface of the armor, such that a substantially equal and opposite force is applied against an impacting projectile,

15 thus reducing the penetrative capability of the impacting projectile.

 A defending force, protecting itself with conventional ballistic projectiles, aim such projectiles by means of sights mounted on the barrel of a gun. Similarly, missiles and other small projectiles are designed to be fired at the attacking target. While various attempts have been made to provide

20 accurate projectiles and missiles, enabling the defending force to fire their weapons while keeping a safe distance from the target, all too often the projectiles reach their target with insufficient velocity to penetrate a vehicle's protective armor. Drag caused by air resistance rapidly reduces the velocity of

a projectile. In order for a projectile to hit the armored vehicle with a velocity sufficient for the projectile to penetrate the target's armor, the defending force must either move closer to the target or wait for the armored vehicle to move closer to them. The reduction of distance, between the defending force and the
5 attacking armored vehicle, exposes the defending force to an ever increasing danger.

Some battle vehicles are so heavily armored, that their armor protects the vehicle's crew from an attack at close proximity. Worse still, modern battle field vehicles often have reactive armor. Even if the modern armored vehicle
10 were to be attacked by a projectile that hits the vehicle's surface with sufficient ability to penetrate its armor, the reactive armor, once triggered, reduces the projectile kinetic energy, preventing any serious damage to the vehicle.

Defending ground forces experience similar problems when encountering armored helicopters and other armored ground attack aircraft.

15 Ground installations are often similarly hardened to protect themselves against attack. Armored installations often house command and control centers operating surface to air installations hostile to aircraft flying overhead. In order to neutralize such a threat, an attacking aircraft launches either free falling ordnance or missiles at the target, only to discover the same problem posed by
20 the tank. Indeed, 'air-strikes' are designed to assist a defending force often prove to be ineffectual against an armored vehicle. The cruise speed of air to surface arms being too low to provide sufficient force to penetrate a target's armor.

There is thus a widely recognized need for, and it would be highly advantageous to have, a long range projectile that impacts its target at penetrating velocity and more particularly, for a high velocity armor piercing shell.

5

SUMMARY OF THE INVENTION

According to the present invention there is provided a projectile for piercing armor. The projectile includes an acceleration rocket motor, for driving the projectile from a cruise velocity to a penetration velocity after the
10 projectile has been launched. The penetration velocity is reached when the projectile impacts with its target. The projectile includes a cruising rocket motor to maintain the projectile's cruise velocity.

According to one embodiment of the present invention, the projectile is a missile.

15 According to another embodiment of the present invention, the projectile is a shell. Preferably the shell is launched from a tank.

According to a preferred embodiment of the present invention, the projectile further includes an armor piercing rod seated within the projectile for piercing armor.

20 According to still further features in the described preferred embodiments, the projectile further includes at least one countermeasure to a reacting target. Preferably the countermeasure includes an advance

projectile associated with the projectile, for neutralizing a target's reactive armor. In one embodiment, the advance projectile is a bullet.

According to another embodiment, the projectile further includes an electronic system to alter the projectile's trajectory during flight.

5 The present invention successfully addresses the shortcomings of the presently known configurations by providing a long range projectile that can strike its target at a sufficiently high speed to penetrate armor.

The present invention discloses a novel method for piercing armor. The method includes the steps of launching a projectile at a target; increasing the
10 projectile's velocity so as to reach a suitable penetration velocity and striking the target with the projectile at the penetrating velocity.

According to one embodiment of the present invention, the method includes the step of maintaining the cruise velocity of the projectile by the cruise motor to reduce deflection of the projectile by side wind, prior to
15 increasing the velocity of the projectile to its impact penetration velocity.

According to one embodiment of the present invention, the method includes penetrating a target's armor with a portion of the projectile, such as by an armor piercing rod seated in the projectile.

According to another embodiment of the present invention, the method
20 further includes employing countermeasures against a reacting target prior to the projectile striking the target.

BRIEF DESCRIPTION OF THE DRAWINGS

The invention is herein described, by way of example only, with reference to the accompanying drawing in which similar reference numbers have been used throughout to designate similar parts, wherein:

5 Figure 1a is a schematic cross-section of a projectile according to one embodiment of the present invention wherein the projectile is a shell;

 Figure 1b is a cross sectional schematic diagram of the projectile of Fig. 1a;

 Figure 1c is a schematic diagram of a shell according to one embodiment
10 of the present invention prior to launch;

 Figure 2 is a schematic diagram of a shell according to a further embodiment of the present invention

 Figure 3 is a schematic diagram of a shell deployed according to one embodiment of the present invention;

15 Figure 4 is a schematic diagram of a missile according to an alternative embodiment of the present invention;

 Figure 5 is a schematic diagram of a missile deployed according to another embodiment of the present invention.

20 DESCRIPTION OF THE PREFERRED EMBODIMENTS

The present invention relates to a projectile that impacts upon its target at a penetrating velocity. The velocity of the projectile is maintained by a cruise rocket motor at a cruise velocity, the speed of the projectile is then

increased by an acceleration rocket motor to a suitable penetrating velocity shortly before impacting upon its target. Specifically, the present invention can be used to provide an armor piercing shell or missile.

For the purposes of the present description and appended claims, a
5 penetrating velocity includes, by way of example only a velocity that allows a projectile, upon impacting a target, to penetrate the target.

An acceleration rocket motor includes, but is not limited to, a rocket propellant, that when ignited, increases the speed of a projectile to a penetrating velocity.

10 A cruise rocket motor includes, but is not limited to a propellant, that when ignited, maintains a cruise velocity of a projectile in flight, while a cruise velocity includes, but is not limited to, substantially any velocity which maintains the projectile's initial launch flight velocity. It will be appreciated that in certain circumstances, a rocket motor can consist only of a rocket
15 propellant.

The principles and operation of a projectile according to the present invention may be better understood with reference to the drawings and the accompanying description.

20 Referring now to the drawings, Figures 1a-1c illustrate a shell 100 constructed according to one embodiment of the present invention. In this embodiment, shell 100 may, by way of example only, be launched from a tank or a cannon.

Shell 100 includes an acceleration propellant 106 annularly concentric to a cruise propellant 116 and an armor piercing rod 104.

Acceleration propellant 106 contained within an inner housing 108 defines an acceleration rocket motor 109. Motor 109 provides a high thrust impulse to shell 100. Propellant 106 can be ignited at a later flight stage of shell 100 and prior to shell 100 impacting its target. In order for maximum acceleration be achieved, from propellant 106, in a short amount of time, it is preferable for propellant 106 to be quick burning.

At least one nozzle 102 is located at one end of shell 100. Nozzle 102 allows hot high pressure gas produced by the burning of propellant 106 to escape. Preferably, nozzle 102 is enclosed within a nozzle housing 110.

Armor piercing rod 104 is seated in a sleeve (not shown) disposed along the vertical axis of missile 100. Rod 104 is preferably long, narrow and sharply shaped to concentrate, upon impacting a target, a penetrating force within as small an area as possible. Rod 104 may be made from a variety of materials including, but not limited to: high strength steel, tungsten alloys, and the like.

Preferably, shell 100 has a multiplicity of stabilizers 114, as shown in Figure 1. Stabilizers 114 increase the aerodynamic stability of shell 100 during flight. Stabilizers 114 preferably deploy once shell 100 has been launched.

As illustrated, shell 100 further includes a propellant 116, located within a second housing 118 annularly concentric to propellant 106, thereby defining

a cruising rocket motor 117. Motor 117 provides shell 100 with an impulse over a relatively long duration of time. Propellant 116 can be ignited either at the launch of missile, or preferably at a latter point in the missiles flight, once shell 100 has reached its cruise velocity. Preferably, propellant 116 is slow burning.

5 Slow burning propellants usually provide a low amount of thrust sufficient to maintain shell 100 at its cruise velocity, increasing the range of shell 100. It is a particular feature of the present invention that cruise motor 117 while maintaining the velocity of shell 100 increases accuracy of shell 100 over larger ranges by minimizing the influence of deflecting vectors such as cross winds.

10 As shown in Figure 1c, shell 100 is coupled by seal 112 to a cartridge 122 containing a launch propellant (not shown) and a primer 126. Primer 126, by way of example only, can be initiated by percussion or electrical current.

Operation of the missile according to the present invention is as follows:

Shell 100 is fired from the gun of a tank, as illustrated in Figure 3.

15 Alternatively, shell 100 can be fired by an artillery gun 338, in the direction of a target 340. Triggered primer 126 causes launch propellant, contained in cartridge 122 to burn, resulting in a sudden increase in pressure in shell 100. The force of the pressure in gun 338 carries 100 out of gun 338 at a muzzle velocity. This explosion also ignites cruise propellant 116 (Figure 1a) of cruise

20 rocket motor 117. The impulse created by motor 117 maintains shell 100 at a cruise velocity, while stabilizers 114 maintain the stability of shell 100.

Prior to shell 100 impacting upon an armored target, impacting upon armored target 340 of Figure 3, acceleration propellant 106 of acceleration

rocket motor 109 is ignited. Propellant 106 may be ignited in any conventional manner, including but not limited to propellant 116 burning its way through housing 108. Alternatively, propellant 106 can either be ignited, at a time pre-set by the weapons operator, by a signal from a proximity sensor located in the front of shell 100, or substantially at the moment shell 100 is launched. Motor 109 increases the velocity of motor 100 to its penetration velocity, thereby enabling shell 100 to strike target 340 of Figure 3 at penetration velocity. The force of shell 100 together with the momentum of rod 104, gained during the flight of rod 104, drive rod 104 into the armor of target 340 until the armor of target 340 is penetrated. Optionally motor 109 can be set to reach an adequate penetration velocity to perforate the target.

Reference is now made to Figure 2, which is a detailed illustrations of a shell 200 constructed and operated according to a further embodiment of the present invention.

In this embodiment, a shell 200 having a cone 240 further includes a communication system having a receiver 230 and a transmitter 232 located in cone region 240 of shell 200. It is an advantage of this configuration that the shell's operator is provided with an opportunity to transmit in-flight instructions to receiver 230 in response to received on-board flight information transmitted by transmitter 232. Optionally, receiver 230 and transmitter 232 can be replaced with a transceiver (not shown), thereby economizing on communication equipment space.

It will be appreciated that a communications system enables the operator to communicate with shell 200, should the operator wish to alter the flight path of shell 200.

Shell 200 also preferably includes an on-board apparatus to neutralize a protective device on targets. As shown in Figure 2, shell 200 further includes a small projectile launching device 234, associated with shell 200, for firing an advance neutralizing projectile 236 at armored targets. Device 234 fires advance projectile 236 either prior to the moment shell 200 hits target 340 of Figure 3 or at the moment shell 200 hits target 340. An advantage of this embodiment is that advance projectile 236 triggers any reactive armor target 340 of Figure 3 may have, thereby leaving target 340 substantially unprotected when shell 200 impacts target 340, thus enabling a greater penetration depth of rod 104.

Operation of the embodiment of Figure 2 is as follows:

As shown in Figure 3, shell 200 is fired, as described above, from a tank gun 338, or from any artillery gun, which by way of example only, may include a 155mm or a howitzer, in the direction of target 340. Shell 200 leaves gun 338 at point "A" having a muzzle velocity. At a point "B", in the flight of shell 200, propellant 116 is ignited, altering the velocity of shell 200 to a cruise velocity. As shell 200 nears target 340, and shell 200 reaches point "C", propellant 106 is ignited at a sufficient distance for enabling the velocity of shell 200 to be altered to substantially a penetrating velocity. Preferably, propellant 106 and propellant 116 are ignited, as described above, at times predetermined by the

operator. Prior to, and at a short distance from, shell 200 impacting target 340, device 234 is triggered to fire projectile 236 at target 340, thus triggering any reactive armor present. Substantially shortly thereafter, rod 104 penetrates the armor of target 340 as described above.

5 Reference is now made to Figure 4, which is a detailed illustration of a projectile constructed according to an alternative embodiment of the present invention. In this alternative embodiment, the projectile is an armor piercing missile 400.

Missile 400 has a cruising rocket motor, generally designated 401, in
10 axial series with an acceleration rocket motor, generally designated 405 and an armor piercing rod 408, located in a sleeve 409 disposed along the vertical axis of missile 400. Rod 408 is also similar to armor piercing rods 104 described in earlier embodiments. Cruise motor 401 includes a cruising propellant 402, located within a housing 410 between a nozzle housing 412 and cruising
15 propellant 402. Motor 401 provides an impulse for propelling missile 400 at a cruising velocity. As shown, a nozzle 414, located within housing 412, is positioned adjacent to propellant 402 to receive hot gases from the combustion of propellant 402. Nozzle 414 directs the flow of hot gases out of acceleration motor 401, thus propelling missile 400 at cruise velocity.

20 Motor 405 is disposed between a compartment 424 and cruise motor 401. Motor 405 includes an acceleration propellant 406, located within housing 416 and a second nozzle housing 418, including at least one nozzle 420. Acceleration propellant 406 is annular shaped having a channel 404. Channel

404 runs down the center of propellant 406. Propellant 406 burns at the center of channel 404 such that channel 404 becomes a combustion chamber providing a larger surface area for propellant 406 to burn. By providing a larger surface area for propellant 406 to burn, a greater volume of hot gases is produced for displacing missile 400 forward at a substantially increased velocity than cruise velocity.

In this embodiment, once motor 401 is spent, motor 401 can be discarded in mid-flight by detaching motor 401 from the rest of missile 400. It is an advantage of this embodiment that missile 400 has less mass being displaced by acceleration velocity by motor 405.

As shown, missile 400 further includes an electronic system 426 located between a guidance system 422 and a sensor 428.

Sensor 428, located adjacent to a sensor dome 430, receives target signals such as a radar signal or heat radiation emitting from targets. Received target signals are then transmitted to electronics system 426. Electronics system 426 processes signals received from sensor 428. These signals are used to calculate the position and distance of target 536 of Figure 5 in relation to missile 400. This information is transmitted to guidance system 422, located in compartment 424, which determines if the trajectory and velocity of missile 400 should be altered as described in earlier embodiments of the present invention.

It is an advantage of the present configuration that information concerning position and distance of target 536 of Figure 5 in relation to missile

400 not only enables the optimal moment to ignite acceleration propellant 406, but also enables the optimal moment of launching of an advance neutralizing projectile 434.

As described in earlier embodiments, missile 400 also includes small
5 projectile 434, to be fired prior to missile 400 hitting target 536, which projectile 536 is disposed within a device 432. Device 432 is disposed between sensor 428 and compartment 424.

Missile 400 can be launched from an aircraft such as an attack aircraft 535, as shown in Figure 5. Alternatively missile 400 can be launched from a
10 ground based platform. Optionally, missile 400 could be fired by a mobile platform, an airborne gunship or a sea going vessel.

Operation of missile 400 is as follows:

As illustrated in Figure 5, missile 400 is released from aircraft 535 at a release velocity, as shown in Figure 5, from an aircraft 535 at a release
15 velocity, substantially contemporaneously with igniting propellant 402 of motor 401 (Figure 4). Motor 401 drives missile 400 from point "A" (Figure 5) to a cruise velocity .

A target which can include by way of example only, a ship, a tank, an artillery station, a radar installation, any ground target, and even an airborne
20 gunship is identified by sensor 428 (Figure 4). Target information is then transmitted to system 426 which transmits updated target location information to guidance system 422. System 422 then determines whether the trajectory of missile 400 should be altered.

As shown in Figure 5, as missile 400 approaches target 536, the optimal distance to target 536, in relation to missile 400, is determined for igniting propellant 406. At this optimal distance marked "B", propellant 406 (Figure 4) is ignited, motor 401 is detached and motor 405 accelerates missile 400 to
5 substantially a penetration velocity.

As described in earlier embodiments, advance neutralizing missile 434 is fired at the target 536 prior to missile 400 impacting target 536, thus neutralizing the reactive armor of target 536. Missile 400 then strikes and penetrates the armor of target 536 as described above.

10 While the invention has been described with respect to a limited number of embodiments, it will be appreciated that many variations, modifications and other applications of the invention may be made.

WHAT IS CLAIMED IS:

1. A projectile for piercing armor comprising:
 - (a) a cruise propellant for maintaining a cruise velocity of said projectile; and
 - (b) an acceleration rocket motor activated after launch for accelerating said projectile from said cruise velocity to a penetration velocity, in a final stage of flight said projectile.
2. The projectile of claim 1, wherein said projectile is a shell.
3. The projectile of claim 1, wherein said projectile is a missile.
4. The projectile of claim 1, further comprising an armor piercing rod situated within said projectile for piercing armor.
5. The projectile of claim 4, for further comprising a device coupled to said projectile for penetrating a reactive target having reactive armor.
6. The projectile according to claim 5, wherein said device includes an advance projectile associated with said projectile, for neutralizing reactive armor of a target.

7. The projectile according to claim 6, wherein said advance projectile is a bullet.
8. The projectile according to claim 7, further comprising an electronic system to alter trajectory of said projectile during flight of said projectile.
9. The projectile according to claim 8, wherein said electronic system further comprising:
 - (a) a sensor, for detecting a target; and
 - (b) a guidance system, for controlling trajectory of said projectile.
10. The projectile according to claim 9, wherein said sensor is responsive to a radar signal.
11. The projectile according to claim 10, wherein said sensor is responsive to radiation emission of said target.
12. The projectile according to any claims 1-11 and substantially described or illustrated herein in Figures 1-5.

13. A projectile for piercing armor substantially described or illustrated herein in Figures 1-5.

14. A method for piercing armor on a target, the method comprising the steps of :

- (a) providing a projectile for piercing armor including:
 - (i) a cruise propellant for maintaining a cruise velocity of said projectile; and
 - (ii) an acceleration rocket motor activated after launch for accelerating said projectile from said cruise velocity to a penetration velocity, in a final stage of flight said projectile;
- (b) launching said projectile at said target;
- (c) maintaining said projectile at said cruise velocity;
- (d) increasing said velocity of said projectile to a penetrating velocity; and
- (e) impacting said target with said projectile at said penetrating velocity.

15. The method of claim 14, further comprising the step of:

(f) penetrating armor of said target substantially subsequently to step (e).

16. The method of claim 15, further comprising the step of:

(g) neutralizing reactive armor of said target prior to step (e).

1/5

FIG. 1A

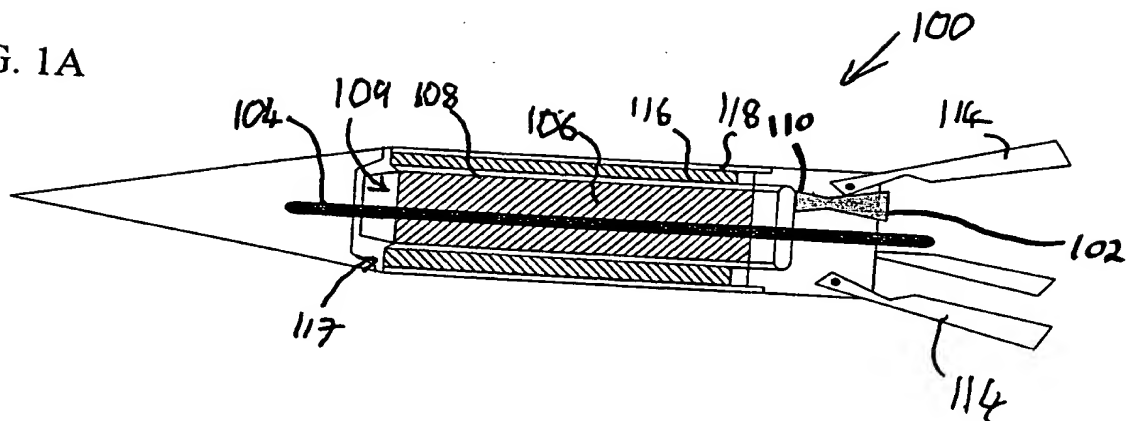


FIG. 1B

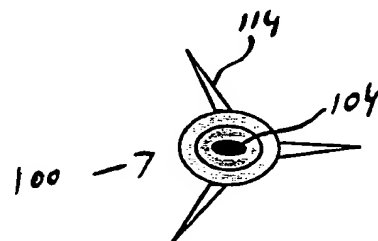
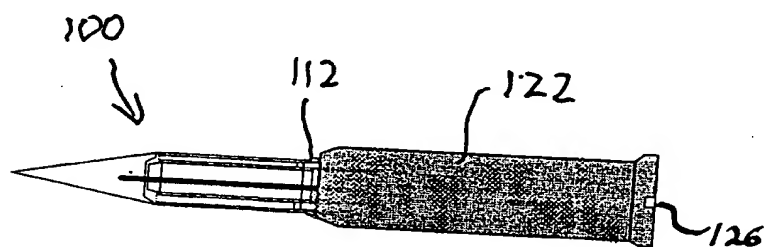
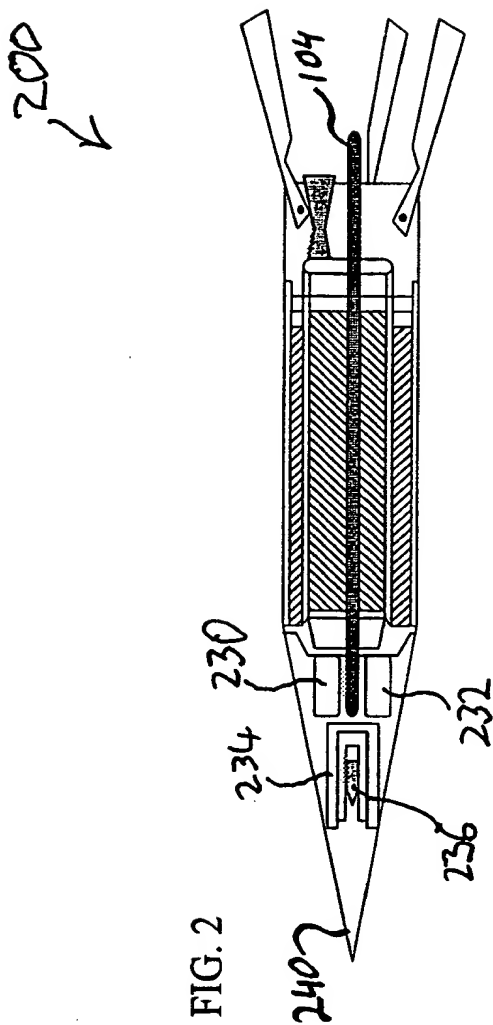


FIG. 1C





3/5

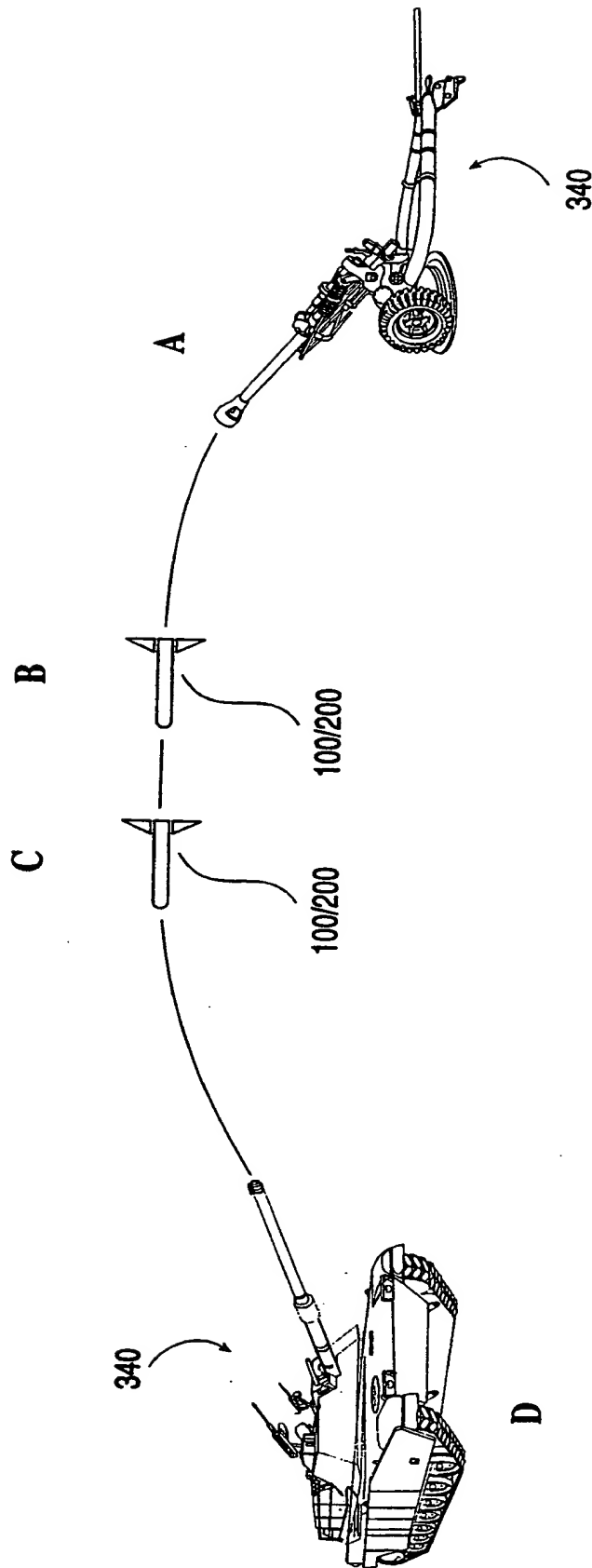


FIG.3

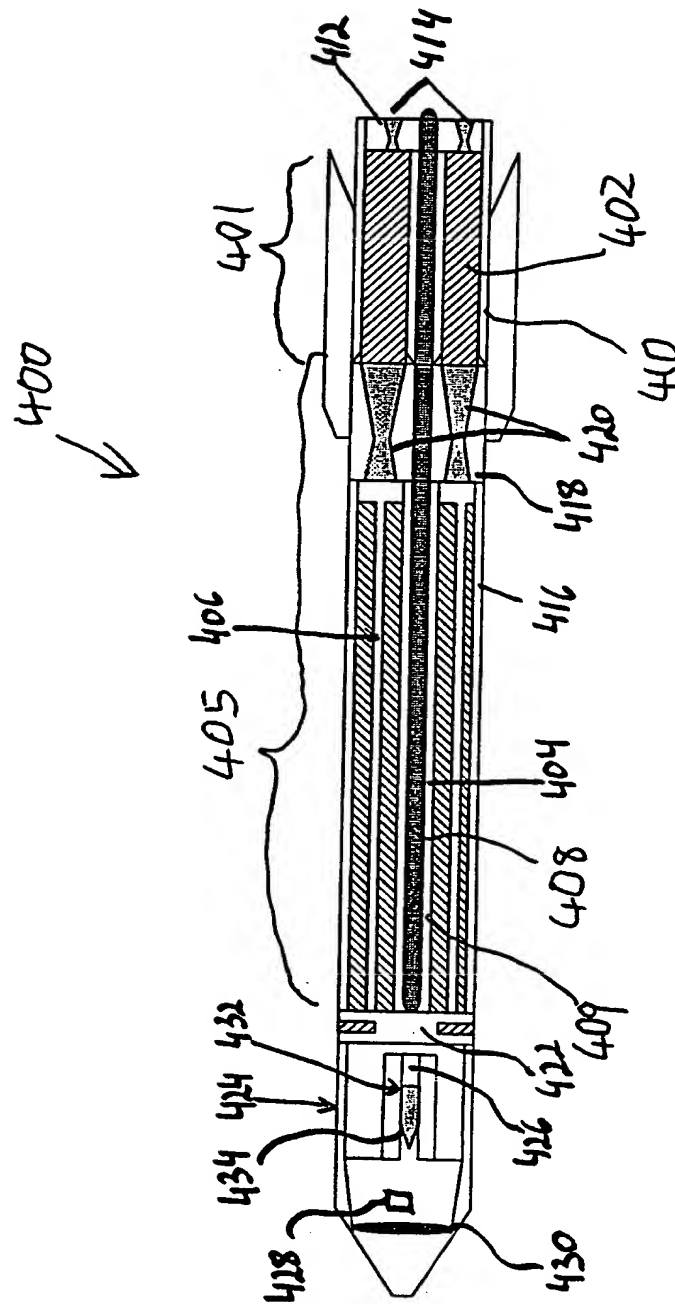


FIG. 4

5/5

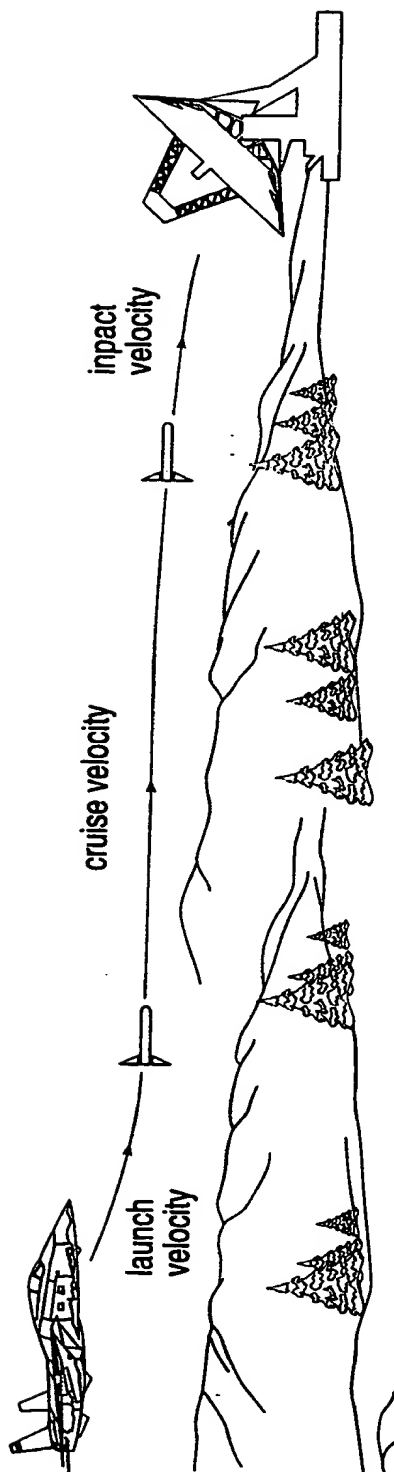


FIG.5

INTERNATIONAL SEARCH REPORT

International application No.
PCT/IL99/00121

A. CLASSIFICATION OF SUBJECT MATTER

IPC(6) : F42B 15/10

US CL : 102/374, 489

According to International Patent Classification (IPC) or to both national classification and IPC

B. FIELDS SEARCHED

Minimum documentation searched (classification system followed by classification symbols)

U.S. : 102/374, 489, 517, 518

Documentation searched other than minimum documentation to the extent that such documents are included in the fields searched
NONEElectronic data base consulted during the international search (name of data base and, where practicable, search terms used)
NONE**C. DOCUMENTS CONSIDERED TO BE RELEVANT**

Category*	Citation of document, with indication, where appropriate, of the relevant passages	Relevant to claim No.
Y	US 4,127,243 A (JACOBSON et al) 28 November 1978 (28/11/78), see col. 2, line 37 - col. 3, line 17.	3,8-11
X --- Y	US 5,189,248 A (DEFFAYET et al) 23 February 1993 (23/02/93), see col. 3, line 41 - col. 9, line 66.	1, 2, 4, 12, 13 ----- 3, 5-11, 14-16
X --- Y	US 5,363,766 A (BRANDON et al) 15 November 1994 (15/11/94), see col. 4, line 1-16.	1, 2, 4, 5, 12-16 ----- 3, 6-11
X --- Y	US 5,596,166 A (GOULD) 21 January 1997 (21/01/97), see col. 4, line 21-31, col. 5, line 52-55.	1, 2, 12, 13 ----- 4-11



Further documents are listed in the continuation of Box C.



See patent family annex.

* Special categories of cited documents:	*T* later document published after the international filing date or priority date and not in conflict with the application but cited to understand the principle or theory underlying the invention
A document defining the general state of the art which is not considered to be of particular relevance	*X* document of particular relevance; the claimed invention cannot be considered novel or cannot be considered to involve an inventive step when the document is taken alone
E earlier document published on or after the international filing date	*Y* document of particular relevance; the claimed invention cannot be considered to involve an inventive step when the document is combined with one or more other such documents, such combination being obvious to a person skilled in the art
L document which may throw doubts on priority claim(s) or which is cited to establish the publication date of another citation or other special reason (as specified)	* & * document member of the same patent family
C document referring to an oral disclosure, use, exhibition or other means	
P document published prior to the international filing date but later than the priority date claimed	

Date of the actual completion of the international search

22 JUNE 1999

Date of mailing of the international search report

23 AUG 1999

Name and mailing address of the ISA/US
Commissioner of Patents and Trademarks
Box PCT
Washington, D.C. 20231

Facsimile No. (703) 305-3230

Authorized officer

THERESA M. WESSON

Telephone No. (703) 308-1113

INTERNATIONAL SEARCH REPORT

 International application No.
 PCT/IL99/00121

C (Continuation). DOCUMENTS CONSIDERED TO BE RELEVANT

Category*	Citation of document, with indication, where appropriate, of the relevant passages	Relevant to claim No.
Y	US 5,728,968 A (BUZZETT et al) 17 March 1998 (17/03/98), see col. 2, line 43-55, col. 3, line 25-53.	4, 5
X	US 5,886,289 A (NIXON et al) 23 March 1999 (23/03/99), see col. 3, line 59 - col. 4, line 62.	1, 2, 12, 13
---		-----
Y		4-11
A	US 3,903,804 A (LUTTRELL et al) 09 September 1975 (09/09/75), see entire document.	1-16
A	US 4,597,333 A (BOCKER et al) 01 July 1986 (01/07/86), see entire document.	1-16
A	US 4,633,782 A (BERUBE et al) 06 January 1987 (06/01/87), see entire document.	1-16
A	US 4,643,099 A (LUTHER et al) 17 February 1987 (17/02/87), see entire document.	1-16
A	US 4,706,569 A (WALLOW et al) 17 November 1987 (17/11/87), see entire document.	1-16
A	US 5,109,774 A (DEFFAYET) 05 May 1992 (05/05/92), see entire document.	1-16

INTERNATIONAL SEARCH REPORT

International application No.
PCT/IL99/00121

A. CLASSIFICATION OF SUBJECT MATTER

IPC(6) : F42B 15/10

US CL : 102/374, 489

According to International Patent Classification (IPC) or to both national classification and IPC

B. FIELDS SEARCHED

Minimum documentation searched (classification system followed by classification symbols)

U.S. : 102/374, 489, 517, 518

Documentation searched other than minimum documentation to the extent that such documents are included in the fields searched
NONE

Electronic data base consulted during the international search (name of data base and, where practicable, search terms used)
NONE

C. DOCUMENTS CONSIDERED TO BE RELEVANT

Category*	Citation of document, with indication, where appropriate, of the relevant passages	Relevant to claim No.
Y	US 4,127,243 A (JACOBSON et al) 28 November 1978 (28/11/78), see col. 2, line 37 - col. 3, line 17.	3,8-11
X --- Y	US 5,189,248 A (DEFFAYET et al) 23 February 1993 (23/02/93), see col. 3, line 41 - col. 9, line 66.	1, 2, 4, 12, 13 ----- 3, 5-11, 14-16
X --- Y	US 5,363,766 A (BRANDON et al) 15 November 1994 (15/11/94), see col. 4, line 1-16.	1, 2, 4, 5, 12-16 ----- 3, 6-11
X --- Y	US 5,596,166 A (GOULD) 21 January 1997 (21/01/97), see col. 4, line 21-31, col. 5, line 52-55.	1, 2, 12, 13 ----- 4-11

☒ Further documents are listed in the continuation of Box C. ☐ See patent family annex.

* Special categories of cited documents:	*T* later document published after the international filing date or priority date and not in conflict with the application but cited to understand the principle or theory underlying the invention
A document defining the general state of the art which is not considered to be of particular relevance	*X* document of particular relevance; the claimed invention cannot be considered novel or cannot be considered to involve an inventive step when the document is taken alone
E earlier document published on or after the international filing date	*Y* document of particular relevance; the claimed invention cannot be considered to involve an inventive step when the document is combined with one or more other such documents, such combination being obvious to a person skilled in the art
L document which may throw doubts on priority claim(s) or which is cited to establish the publication date of another citation or other special reason (as specified)	*G* document member of the same patent family
O document referring to an oral disclosure, use, exhibition or other means	
P document published prior to the international filing date but later than the priority date claimed	

Date of the actual completion of the international search

22 JUNE 1999

Date of mailing of the international search report

23 AUG 1999

Name and mailing address of the ISA/US
Commissioner of Patents and Trademarks
Box PCT
Washington, D.C. 20231

Facsimile No. (703) 305-3230

Authorized officer

THERESA M. WESSON

Telephone No. (703) 308-1113

INTERNATIONAL SEARCH REPORT

 International application No.
 PCT/IL99/00121

C (Continuation). DOCUMENTS CONSIDERED TO BE RELEVANT

Category*	Citation of document, with indication, where appropriate, of the relevant passages	Relevant to claim No.
Y	US 5,728,968 A (BUZZETT et al) 17 March 1998 (17/03/98), see col. 2, line 43-55, col. 3, line 25-53.	4, 5
X	US 5,886,289 A (NIXON et al) 23 March 1999 (23/03/99), see col. 3, line 59 - col. 4, line 62.	1, 2, 12, 13.
---		-----
Y		4-11
A	US 3,903,804 A (LUTTRELL et al) 09 September 1975 (09/09/75), see entire document.	1-16
A	US 4,597,333 A (BOCKER et al) 01 July 1986 (01/07/86), see entire document.	1-16
A	US 4,633,782 A (BERUBE et al) 06 January 1987 (06/01/87), see entire document.	1-16
A	US 4,643,099 A (LUTHER et al) 17 February 1987 (17/02/87), see entire document.	1-16
A	US 4,706,569 A (WALLOW et al) 17 November 1987 (17/11/87), see entire document.	1-16
A	US 5,109,774 A (DEFFAYET) 05 May 1992 (05/05/92), see entire document.	1-16